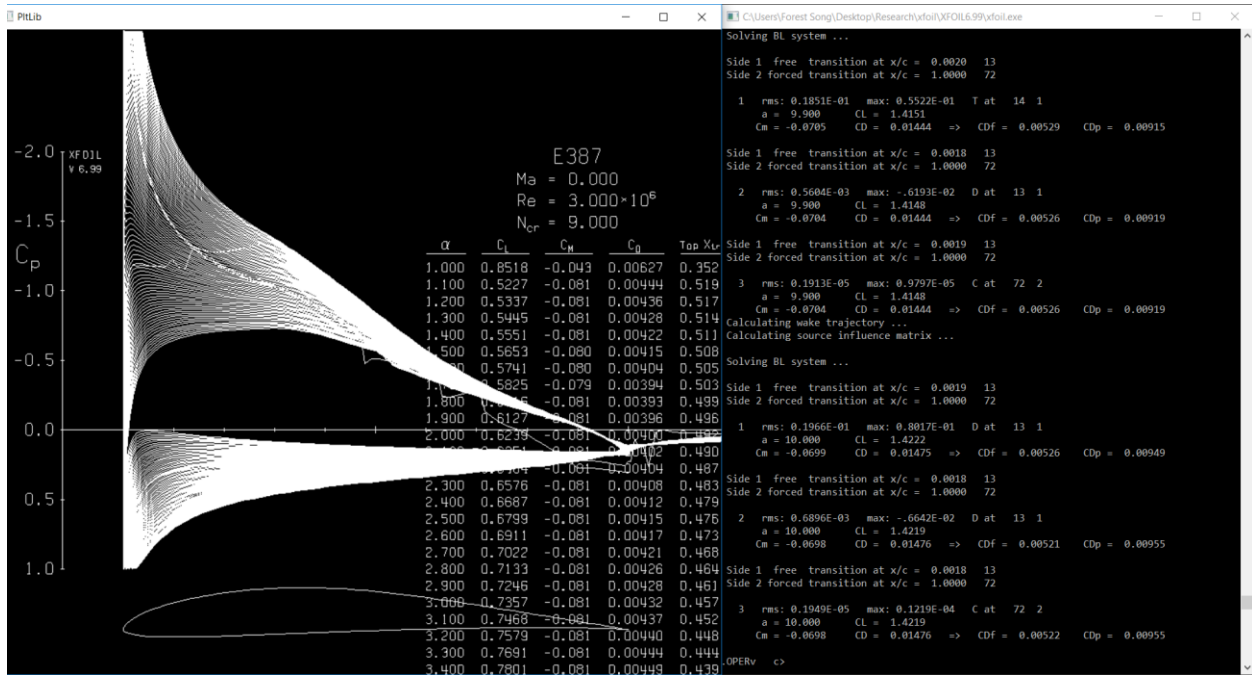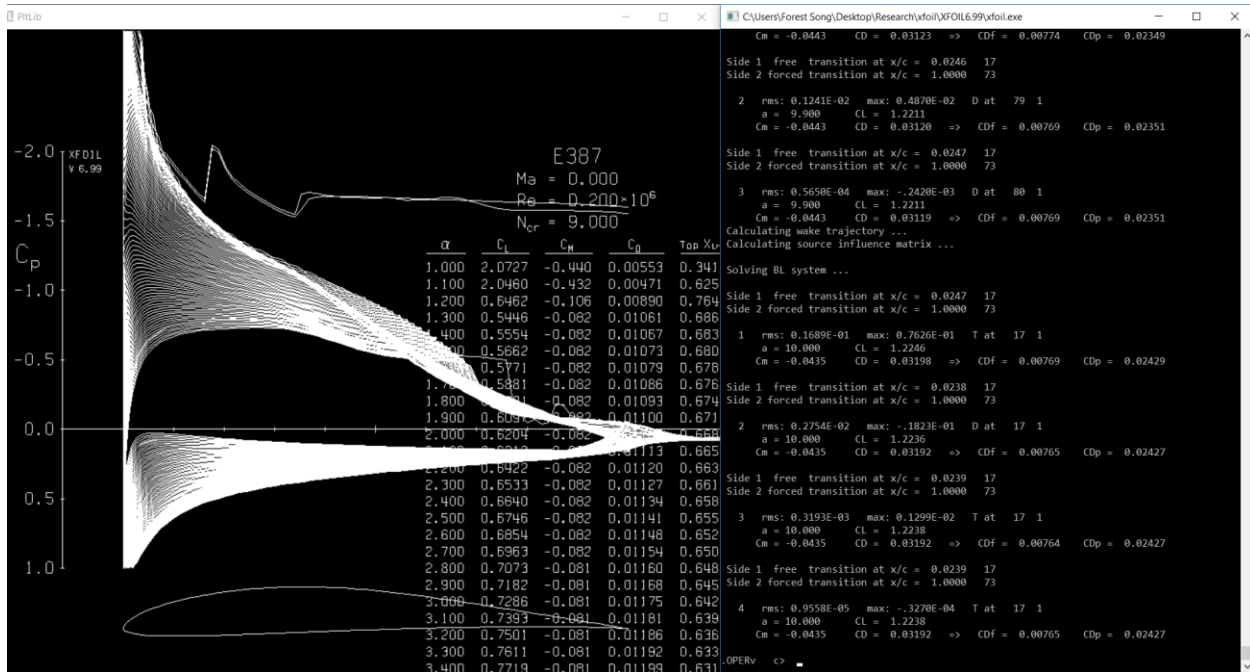Reynold's # = 1e5, changing angle of attack(alpha) from 1-10 with .1 degree increments, converged up until alpha = 8.6

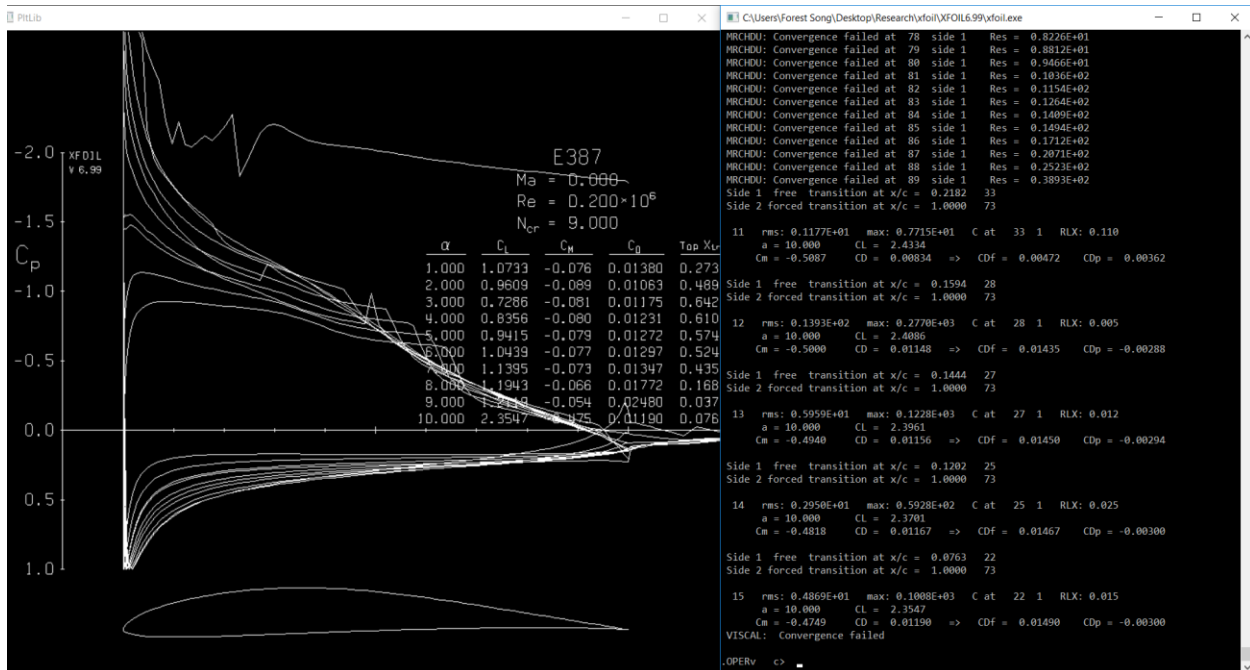Reynold's # = 3e6, changing angle of attack(alpha) from 1-10 with .1 degree increments, converged for all values of alpha

E387

Ma = 0.000
Re = 0.200×10⁶
Ncr = 9.000

α	CL	CM	CD	Top Xtr
1.000	2.0727	-0.440	0.00553	0.341
1.100	2.0460	-0.432	0.00471	0.625
1.200	0.6462	-0.106	0.00890	0.764
1.300	0.5446	-0.082	0.01061	0.686
1.400	0.5554	-0.082	0.01067	0.683
1.500	0.5662	-0.082	0.01073	0.680
1.600	0.5771	-0.082	0.01079	0.678
1.700	0.5881	-0.082	0.01086	0.676
1.800	0.5991	-0.082	0.01093	0.674
1.900	0.6097	-0.082	0.01100	0.671
2.000	0.6204	-0.082		
2.200	0.6422	-0.082	0.01113	0.665
		-0.082	0.01120	0.663
2.300	0.6533	-0.082	0.01127	0.661
2.400	0.6640	-0.082	0.01134	0.658
2.500	0.6746	-0.082	0.01141	0.655
2.600	0.6854	-0.082	0.01148	0.652
2.700	0.6963	-0.082	0.01154	0.650
2.800	0.7073	-0.081	0.01160	0.648
2.900	0.7182	-0.081	0.01168	0.645
3.000	0.7286	-0.081	0.01175	0.642
3.100	0.7393	-0.081	0.01181	0.639
3.200	0.7501	-0.081	0.01186	0.636
3.300	0.7611	-0.081	0.01192	0.633
3.400	0.7719	-0.081	0.01199	0.631

Reynold's # 2e5, changing angle of attack(alpha) from 1-10 with .1 degree increments, converged for all values of alpha

Therefore, increasing the Reynold's # will allow for greater chance of convergence

Lower pressure above the wing and high pressure below the wing will generate lift, explains the coordinates of the x axis for Cp.

E387

Ma = 0.000
Re = 0.200×10⁶
Ncr = 9.000

α	CL	CM	CD	Top Xtr
1.000	1.0733	-0.076	0.01380	0.273
2.000	0.9609	-0.089	0.01063	0.489
3.000	0.7286	-0.081	0.01175	0.642
4.000	0.8356	-0.080	0.01231	0.610
5.000	0.9415	-0.079	0.01272	0.574
6.000	1.0439	-0.077	0.01297	0.524
7.000	1.1395	-0.073	0.01347	0.435
8.000	1.1943	-0.066	0.01772	0.168
9.000		-0.054	0.02480	0.037
10.000	2.3547	-0.075	0.01190	0.076

Changing the angle of attack with only 1 degree increments will cause it to not converge.